

What is claimed is:

1. A high-pressure turbine of a gas-turbine engine comprising: a turbine disk carrying rotor blades and rotor blade platforms, a stator ring carrying stator blades and stator blade platforms, a lateral wheel cavity formed between the turbine disk and the stator ring, and a seal provided in an axial direction between the stator blade platforms and the rotor blade platforms which is arranged radially outwardly from a center axis of the high-pressure turbine and adjacent a main gas duct.
2. A high-pressure turbine in accordance with Claim 1, wherein the lateral wheel cavity is a single cavity.
3. A high-pressure turbine in accordance with Claim 2, wherein the rotor blade platforms form a seal runner.
4. A high-pressure turbine in accordance with Claim 3, wherein the seal is a segmented labyrinth seal with labyrinth tips positioned on the blade platforms.
5. A high-pressure turbine in accordance with Claim 2, wherein the seal is a labyrinth seal, with labyrinth tips attached to the stator ring and with the rotor blade platforms forming a segmented seal runner.
6. A high-pressure turbine in accordance with Claim 2, wherein the seal is of a brush type, with brush elements attached to the stator ring and with the rotor blade platforms forming a segmented seal runner.
7. A high-pressure turbine in accordance with Claim 2, wherein the seal is of a brush type, with individual brush elements positioned on the rotor blade platforms.
8. A high-pressure turbine in accordance with Claim 1, wherein the rotor blade platforms form a seal runner.

9. A high-pressure turbine in accordance with Claim 1, wherein the seal is a segmented labyrinth seal with labyrinth tips positioned on the blade platforms.
10. A high-pressure turbine in accordance with Claim 1, wherein the seal is a labyrinth seal, with labyrinth tips attached to the stator ring and with the rotor
5 blade platforms forming a segmented seal runner.
11. A high-pressure turbine in accordance with Claim 1, wherein the seal is of a brush type, with brush elements attached to the stator ring and with the rotor blade platforms forming a segmented seal runner.
12. A high-pressure turbine in accordance with Claim 1, wherein the seal is of a
10 brush type, with individual brush elements positioned on the rotor blade platforms.
13. A sealing arrangement for a high-pressure turbine of a gas-turbine engine having a turbine disk carrying rotor blades and rotor blade platforms, a stator ring carrying stator blades and stator blade platforms and a lateral wheel
15 cavity formed between the turbine disk and the stator ring, the sealing arrangement comprising a seal provided in an axial direction between the stator blade platforms and the rotor blade platforms which is arranged radially outwardly from a center axis of the high-pressure turbine and adjacent a main gas duct.